

77-19-07 BEECH: Amendment 39-3042. Applies to all Beech 18 series airplanes, including all military counterparts thereof and those Beech 18 series airplanes modified in accordance with supplemental type certificates.

COMPLIANCE: Required as indicated unless already accomplished as specified below.

To detect any cracking of the pilot's and copilot's control column and the elevator control arm assembly, within 30 days after the effective date of this AD and each 300 hours time-in-service or 12 calendar months thereafter, whichever occurs first, accomplish the following in accordance with Beechcraft Service Instructions No. 0889-155 or later approved revisions and in accordance with the additional requirements of Paragraph A. below:

- A. Referring to Figure 1 of this AD inspect the areas identified as "A" and "B" on the pilot's and copilot's control column assembly for cracks using visual and either magnetic particle, or penetrant procedures. Repair or replace any cracked control column components, using airworthy parts, prior to the next flight.
- B. Referring to Figure 1 of this AD inspect the Beech P/N 187504 elevator control arm assembly for cracks using penetrant procedures. Special attention should be given to the area around the taper pins. Replace any cracked P/N 187504 elevator control arm with an airworthy component prior to the next flight.
- C. Within one week after any crack is found during the above-noted inspections, submit a written report to the FAA via an FAA M or D Report (FAA Form 8330-2) or a letter to Chief, Engineering and Manufacturing Branch (ACE-210), FAA, 601 East 12th Street, Kansas City, Missouri 64106. This report must include identification of the part that is cracked, the length and location of each crack, hours total time-in-service on the cracked component, and the aircraft serial number. (Reporting approved by Office of Management and Budget under OMB No. 04-R0174.)
- D. Aircraft may be flown in accordance with FAR 21.197 to a base where this AD can be accomplished.
- E. The interval for the repetitive inspections set forth in this AD may be extended 30 hours up to a maximum of 330 hours time-in-service or, if applicable, one calendar month up to a maximum of 13 calendar months to allow compliance at previously scheduled maintenance periods.
- F. Any equivalent method of compliance with this AD must be approved by the Chief, Engineering and Manufacturing Branch, FAA, Central Region.

This amendment becomes effective on September 29, 1977.

